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Aerial View of the Statue of Liberty on Bedloe's Island

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Number 10

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No. 10

British Admiralty on Aircraft

THE First Lord of the Admiralty in explaining the British naval estimates for 1920-1921 gave as his opinion that the capital ship was not doomed by aircraft. He is quoted as saying: "We at present would the Board of Admiralty subscribe to the statement that aircraft have caused the capital ship Aircraft are not a substitute for the capital ship. Aircraft are not a substitute for the capital ship. Aircraft are not a substitute for the capital ship."

This statement was made some time ago, but it is significant that reports from London indicate that the British capital ship program has been temporarily halted.

The Significance of Scurvy

FOR the last two years it has been generally assumed that the United States had complete information of what progress was being made in other countries in aircraft development. Much was written regarding the great improvements that had been made in construction and speed.

Suddenly, last year, the channels of information seemed to close and now it is almost impossible to learn what the more advanced of our former allies are doing in the progress of their air services. The prime question of aircraft carriers is a specific instance. Our information on foreign ships of this type is meager and little seems to be forthcoming from the usual channels. Such a condition will give rise to difficulties which may cause the cry for disarmament to take on a new tone. If nations are willing to have their army and navy programs known, but withhold at the same time their air program, the conclusion must be reached that it is in this branch of armament that the most significant developments are taking place.

Length of Tail Surface Area in Monoplanes

In standard biplane design a rule which is fairly well established is to make the area of the tail surface or the distance from the center of the horizontal tail surface to the center of gravity of the machine about three times the length of the wing chord.

When a monoplane design is employed, extensive gain is secured—with resulting increase in weight of wing construction and considerations—by cutting down the aspect ratio from the customary seven or six to a value as low as four. The chord is thereby largely increased. If the horizontal tail surface area is given the same percentage of the wing area as in biplane practice, it would mean, at first sight, legend to increase the distance from the tail surface center to the center of gravity in proportion to the increase in chord. This would involve lengthening of the fuselage which means again increase in weight of the fuselage and increase in overall dimensions. What is at first logical is therefore not desirable.

It should also be remembered that if the chord increases in the monoplane very largely, the moment of inertia does not increase in anything like the same proportion. This has a very important bearing on the problem.

If the tail surface area of the long chord monoplane were made three times the chord length, the degree of static stability would result what it is for the ordinary biplane, but the relative damping effect in dynamic conditions would be very greatly increased. Since the damping in an ordinary biplane is ample, this increase in relative damping effect would be superfluous, if not positively undesirable.

It is not suggested that in view of these preliminary considerations, the designer of the monoplane should at once make a radical reduction in proportioning the tail area, but it is believed that a further consideration of the problem of reducing the tail area might be worth while.

Airline Factors of Safety

In considering the safety of the tail surfaces of an airplane, arbitrary values are generally set for the loads per square foot to be imposed. Sometimes consideration is given to the maximum tail loads possible with reference to the center of pressure movement on the wings. In considering the safety of the airplane, it would seem that something better than the arbitrary setting of a certain load per square foot is possible.

The loads which can be imposed on the airplane depend to a certain extent on the pilot's strength. But in a small machine with the narrow chord surfaces of a monoplane, the pilot might never have to exert himself greatly to obtain maximum displacement of the airplane. The question then arises as to what the effective loads would indeed be related to. They would certainly be related to the loading of the main surface and also the factors of safety of the main surface. The logical process would be to agree as follows: the normal wing loading is so much, the factor of safety for the wings is so much; under a given maneuver condition the loading on the wings approaches the product of these two values. When the wings are under this heavy load, a maximum displacement of the airplane imposes a greater load on the wings. The airplane therefore has a loading, which from pressure distribution experiments, we have been a certain rule to the loading on the wing surface. In front of it. From these considerations a fairly sound rule might be devised, relating the effective factors of safety to the design of the ship as a whole.

Airship Mooring Masts

THREE mooring mast which is illustrated in this issue is an important development which has proved successful. By its use the cost of airship operation will be considerably reduced and greater safety will be achieved.

The Night Observation Airplane U.S.X.B.1-A

The ghost of the Bristol Fighter is seen in the U.S. EB-1-A, the new two-seater night observation airplane shortly to be sent to units of the Army Air Service along the Mexican border. This machine is similar in the day type in all respects, save that it is fitted with landing flares, lighting system and other equipment for night flying. A machine for this work will eventually be altered and painted to be as nearly invisible as possible.

The X-B1-A is, in fact, a development of the Draco Fighter, which was to have been fitted with a Liberty and put into production in 1917 for overseas use. Changes have been made from time to time during the intervening period until only the general outline remains. For instance, it has



Fig. 5. View of the XB1A Night-Oscillation Airplane

a vanner body and a 300 hp Wright engine, which is better than the Liberty for high altitude work where a booster is not fitted.

The machines are being tested to show 125 m.p.h. at ground level, and 129 m.p.h. at 5000 ft., down to 83 m.p.h. at 22,490 ft. Altitude, the X-51 has a speed of about 110 m.p.h. at 5000 ft., and of 160 to 200 m.p.h. above 10,000 ft. Compared with the D-155 the machine on which production for commerce centered during the World War, the X-51 A has the advantage of about 35 m.p.h. more speed at the ground and 9 p.s.f. at 5000 ft. The X-51 A stands to 15,000 ft. in 10 sec., compared with the Bristol Blenheim of 35 sec., 30 sec. to 12,000 ft., and the D-15 A of 24 sec.

It will be at high altitude that the pilot will see a big improvement on the D-11-E-1. From 15,000 ft. upward it will be found much faster, much more maneuverable, much lighter on controls and altogether more like a single-engine fighter than any other phase of this type heretofore produced or used by the U. S. Army Air Service.

The flying qualities have been found to be exceptionally good at all altitudes. The airplane is neither nose heavy nor tail heavy, due to the adjustable stabilizer, which is very effective. The airplane can be flown with hands off; it will also slide with hands and feet off controls.

The landing qualities are also very good. It requires about 600 ft. to complete a landing in calm air. There is no tendency to nose over, and it taxis very well, though the rudder is small and the tail drift is not abnormally small.

Verdilly is exceptionally good ahead and above. Pilot and gunner are both able to look ahead, above and below the top wing. The lower wing, of course, as in all airplanes of

this type, is objectionable. The locations of the controls in general are very good.

Discussion

Overall span, 36 ft., 4 9/16 in.; over-all length, 26 ft., 6 in.; over-all height, 6 ft., 9 1/2 in.; height at hub of propeller above ground in flying position, 5 ft., 7 in.; at rest, 6 ft., 7 in.

4748

Wing-curve, modified R A F 15; sweepback, none; dihedral, 34°; drag, stagger, 16 in.; total area, including ailerons, 665 sq. ft.; max. 5 ft. 5 in.

Maximum cross-section shape, largest crosswise, maximum cross-section area, 8.3 sq ft.; maximum cross-section diameter, 32 ft. by 45 in.; distance of maximum section from leading edge, lower chord, 26 in.

Landing Gear

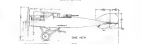
Number of wheels, 2; tread, 66 in.; shock-absorbing system, rubber; braking device, none; wheel stand of 11 in.; 125 in.

File

Area: 10.9 sq. ft.

Power Plant

One Wright, model II, 300 kg., at 1800 rpm., weight, 0.022 lb.; gas consumption, 0.021 lb. per hp. hr.; oil consumption, 0.005 lb. per hp. hr.; weight of water in engine

OUTLINE DRAWING OF FIG. 3 B 1-A. NUCLEO-GLOSSOLINOL
ANALOG[illegible]

Distribution of Weight
 Weight empty (vial, water, 2,600 lb.; instrument, and equipment, 283 lb.); crew, 360 lb.; 300 lb. of gear.
 Weight loaded, 2,994 lb. Weight on front wheels (tail skid on ground), 2,820 lb.; weight on tail skid (tail skid on ground), 364 lb.; weight on front wheels (flying position), 2,793 lb.; weight on tail skid (flying position), 284 lb. Center of gravity (distance from wheels in flying position), 3.15 ft. Weight, 2,994 lb.; wing area, 170 sq. ft.; wing loading, 17.6 lb./sq. ft. (empty), 19.3 lb./sq. ft. (loaded). Stall speed, 50,000 ft./min. (empty), 55,000 ft./min. (loaded). Maximum speed at sea level, 60,000 ft./min. (empty), 65,000 ft./min. (loaded).

Aeronautical Engine Testing Stations

Laboratories for the testing of aeronautical engines, both of old and new design, are located at Dayton, Ohio, for the Army; at the Washington Navy Yard, Washington, D. C., for the Navy; and at Langley Field, Virginia, for the National Advisory Commission for Aeronautics. Upon the recommendation of the Army, Navy or the National Advisory Commission for Aeronautics, the Bureau of Standards, Washington, D. C., will carry out tests of engines and accessories with special reference to their performance under altitude conditions, in a special laboratory designed for this purpose.

Inventions and new appliances applicable to aeronautical engines may be submitted to the National Advisory Committee for Aeronautics, Washington, D. C., whose Committee on Power Plants for Aircraft may arrange for test by the Government of engines or devices which it may approve.

Unless a designer or inventor can furnish facts of interest showing that his proposed engine has passed the research stage and is of moderate value for use in military or commercial aircraft, the Government will not test the engine at its expense. On the other hand, if an engine shows promise of development, either the Army or the Navy will bear the expense of test and of further development.

It is not the policy of the Laboratories mentioned to make tests of aeronautical engines for any one who may offer them for that purpose, but to do so only when in their judgment the engine is worthy, or on specific recommendations for the N.A.A.A.

Aircraft engines may be submitted to the Director of Air Service, Washington, D. C., or to the Navy through the Engineering Chief, Chief Bureau of Steam Engineering, Navy Building, Washington, D. C.

Inventions and unpatented new appliances applicable to aeronautical engines submitted to any War Department office must be forwarded to the Inventions Section, Operations Division, General Staff, State, War and Navy Building, Washington, D. C.

There are no government funds available for aiding new enterprises or developing private investments.

Aviation School Expands

As evidence of the growth of interest in flying on the Falls to Canal is the fact that the Dredney Aircraft Corporation, which for the past year has been conducting ground and flying courses at Portland, Oregon recently completed a five year lease of a building at 105-107 North 11th St. which will soon house

Shops have been established so that the shop and lecture work of the school can all be done in the new headquarters which has quarters for three school.

The school, which has been in existence less than a year already, has about eighty students, and this number is being rapidly augmented, according to an announcement of the trustees.

Officials of the school have announced that additions to the shop and other equipment of the school are to be made immediately, looking to making it the most modern aviation school west of the Mississippi river. This will include purchases of machines, airplane parts and airplanes.

Fuel Feed Systems for Airplanes

By L. B. Lent

One of the outstanding lessons learned in over two and a half year's operation of the Air Mail Service is that altogether too much trouble is encountered in the operation and maintenance of fuel feed systems. The constant source of trouble is not peculiar to any type of plane nor is any particular type of engine, but is found in a more or less degree in them all. It is a fact to say that most of the forced landings, not due to bad weather, and a large part of the fuel maintenance work is caused by trouble with the fuel feed system. If all of it could be eliminated, the work and worry of field crews would be much reduced. There should be no more trouble with the fuel system in a plane than there is in an automobile, and it is felt that if this phase of airplane engineering is given the amount of effort it deserves we could almost forget the system of the fuel system, as we now do in our automobiles. It is unnecessary to point out a faulty system, a potential fire hazard and the possible cause of fatal accidents. This subject merits the best thought that can be given to it.

We will discuss this matter briefly, pointing out some of the features developed by the Air Mail Service, and indicate some lines along which improvement may be accomplished.

Up to the present time there has been no endeavor to standardize fuel systems for airplanes nor their various parts. Each manufacturer seems to have developed a system of his own and thus there are about as many fuel feed systems as there are makes of planes. This is true even of planes with a single engine power plant. When multi-engine planes are considered, the many levers and the mass of pipe and connections and crossovers are so complex that they require explanation as well as they do. But, it also has caused many designers and many more mechanics who have to keep this complex system of "bushings" in order, to give serious thought to developing some system which is more simple, reliable and fuel proof, for it requires an argument to prove that the simpler the system, the less the trouble in the field.

For the purpose of discussion, the subject may naturally be divided into two parts, that of fuel storage and that of fuel delivery. The first deals almost entirely with storage tanks and the latter with pump systems, connections, valves, strainers, etc., which comprise the delivery system.

Fuel Storage Tanks

Theoretically the strongest shape tank would be one of spherical shape, but it is difficult to manufacture such a shape and other important considerations preclude its use. Next to a spherical tank, a cylindrical one with hemmed heads, such as constitutes the shell of submarine steel tanks, possesses the greatest strength per unit of weight.

The consideration which seems to have determined the shape of most tanks in the Air Mail Service is that they get them. They have been so shaped as to fill or empty as far as possible the available space in the fuselage. Tanks placed in, or on, or under wings have, of course, been so shaped as to fill in little tank pockets thus forcing the fuselage to be made, however, the result is a sort of shape with one or more large flat surfaces. Such surfaces must be well stayed with internal plates or bracing, especially if the delivery system uses pressure to prevent the fuel from flowing out of the system. This pressure added to the inertia forces due to the surge of the fuel tends more than most designers imagine and results in more damage than is commonly supposed. Many are likely surprised to find that the Air Mail Service has had their fuel tanks all bulged out between the ribs and leak badly at both the ribs and seams.

It should be unnecessary to mention the fact that a gas tank with its fittings should be made strong to stand any and all wear and tear of ordinary use, but the many failures seem to indicate that mechanical strength is too often sacrificed to saving weight. Bracing and reinforcing are frequently relied upon to carry stresses instead of putting all most stresses on rivets or other fastenings and only relying on them to make the joint gas tight.

It may seem almost foolish to question such obvious wisdom, but the experience of the Air Mail Service shows that many tanks are structurally weak, require much repair and maintenance and hence furnish work for crews which should not be necessary. And it is the sum total of all these little unnecessary jobs which so increases maintenance work and makes life miserable for the boys on the ground.

When the amount of gas to be stored is comparatively small, say fifty gallons or more, it becomes a question as to the use of one or more tanks. One large tank ordinarily weighs less than two or more smaller ones of the same total capacity, although the larger tank requires thicker plate for the reinforcement of the dome. A consideration which might be dangerous is that a leak in one of two or more tanks is no more serious than a leak in one of two or more tanks in a single tank, but this applies only when each

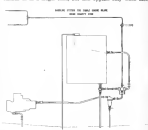


Fig. 1

tank is equipped with a shut-off cock or valve. This tends to complicate rather than simplify, a thing to be avoided if possible.

It is the writer's opinion that when other conditions do not accommodate the use of two or more tanks, a single tank of the requisite capacity is preferable. Inasmuch as we are discussing commercial machines principally, the matter of leak-proof, leak-proof and crash-proof tanks can well be left to a consideration of military machines.

A properly built tank should be leak-proof, but it should be borne in mind that at the present time they are not so built and in many cases it might be advisable to use the so-called leak proof and crash-proof tank. Several of the rubber manufacturing companies have developed very satisfactory rubber blankets or screens which do not add prohibitive weight but which tend to render a gas tank both leak-proof and crash-proof to an extraordinary degree.

Gasoline Distribution Systems

Before beginning a detailed discussion of this part of the subject, let us agree and finally do in our minds the phrase that most important fact that the one object of all delivery systems is to furnish to the one or more carburetor heads of the power plant a proper quantity of clean, pure gas (preferably at a constant pre-determined pressure) under all conditions of flight. This, of course, means all conditions of

angle loading and all portions of the plane in flight. But a most important consideration is that this must be accomplished while reducing chance of leakage to an absolute minimum, thus lowering or eliminating the fire hazard.

One of the main, but inferior, causes of leakage in the distribution of the structure in which the storage and delivery system are installed. An airplane is by no means rigid and is subject to many or less vibrations and this in turn induces stress in the piping system carried thence which are sometimes very severe. It is these vibration stresses which are the prime cause of leakage and rupture, and both occur mostly at joints and connections, so that for airplanes work there have been developed types of connections adapted to the work, but which at present are still the source of much annoyance and trouble and sometimes of fatal accidents. These leaks in gas

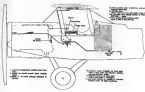


Fig. 2

line may result in loss of life is sufficient reason why every possible effort should be made to eliminate this source of accident.

Definition of Distribution Systems

The several methods of distributing gas to one or more carburetors of the power plant may be classified as the gravity system, the air pressure system and pump system. Some systems use a combination of two of them in the same plane.

It is presumed that the reader is familiar with them, but it may be well to state that the pure gravity system relies on a difference in relative elevations between the tank and the carburetor head to induce a sufficient flow of gas as in the De Havilland machine.

The air pressure system employs an air pump, driven by the engine, to create sufficient air pressure in a closed tank to drive gas to the carburetor. Such a system is used in the De Havilland machine. In the pump system, some form of pump is used to force the liquid through the system.

Gravity Systems

The gravity system is ideal if a sufficient head can at all times be maintained and kept within certain pressure limits, but in pump systems this is quite impossible. The head eventually diminishes as any gravity tank is emptied. Furthermore, there are many cases in which it is impossible to install a tank in the main tank in such a position as to supply sufficient head. For instance in most Liberty engine installations the carburetor is in the valley between the cylinders so high in the fuselage to flow gas to it by gravity from any possible main tank location. Hence, in some Army De Havilland planes, an engine driven air pump supplies air pressure to the gas tank. The Martin and the Curtiss machines, on the other hand, carburetor gas tanks are mounted over each engine in the upper wings and gas pumped to them from the main tank located in the fuselage and tank to the carburetors. This is a typical example of the combination of the pump and gravity system. An overflow is of course provided from the gravity to the main tanks to return the excess gas pumped.

The advantage of the gravity system when it is possible to use it is simplicity of piping system and connections, fewer fuel tanks proper strainers and little danger of failure, due to simplicity.

The disadvantages are, constantly decreasing head as gas tank empties and liability to secondary leak of gas, due to inertia of gas in pipe lines caused by the motion of the plane, such as in lumpy air or in some maneuvers.

Pressure Systems

It should be unnecessary to discuss at length the use of air pressure to drive gas out of the tank and force it through the system to the carburetor, for the system at best is working but has no means to an end and has little to commend it except its simplicity in certain cases. The proper operation is based on

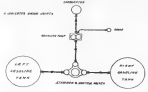


Fig. 3

an over-pressured, tight air line and on the proper functioning of a spring-loaded air discharge valve for returning all excess gas pumped and on maintaining a constant air pressure on the tank, and finally upon the continuous operation of an engine-driven air pump, which however is usually quite reliable.

This system is used in many De Havilland planes in the Air Mail Service and is a source of much annoyance and trouble. Air flow and tanks are difficult to keep tight and the air relief valves are a source of much trouble and need much attention. There is also a considerable system of air and gas in the gas tank. One of the main faults in this system has served its purpose and will soon go into the discard.

Positive Gas Pump Systems

And now let us consider briefly the positive gas pump system, which is the writer's opinion deserves some consideration and thought thus appears to have been put on it in all its simplicity. It is obvious that, in order to obtain a sufficient quantity of gas under proper pressure at the carburetor, some method of controlling the output of the pump or of returning the excess gas pumped to the main tank must be used. The two systems most used are: A secondary gravity tank which is placed at sufficient height above the carburetor to furnish the necessary head, with an overflow from this tank back to the main tank to return the excess gas pumped; or some method of returning the excess gas pumped through a return line containing a spring-loaded check valve which can be set to open at a pre-determined pressure and so maintain the pressure at the carburetor and also return to the main tank or tank the excess gas over that being consumed by the engine.

Types of Pumps

The system has suffered in the past from some lack of simplicity, but none from unreliable pumps. About all types have been used, viz, plungers, gear and centrifugal and in many American machines they have been used by some form of wind power, instead of mechanical power or electricity. A pump driven by the engine is much better in all respects. First, because it is positive and runs at a speed corresponding

to the engine's demand for gas, and second, because a proper form of drive puts less strain on the pump than in the present form of wind-driven pump.

In most wind-driven pumps the speed is much greater than any proper pump speed (except the centrifugal pump) and accumulates intermediate starting when engine pump speeds are in its intervals and need. Furthermore, it is almost impossible to control the speed of a wind-driven pump. When

it fails to keep the shaft properly packed against leaks. For these and other reasons, the engine drive is more reliable and satisfactory.

Much might be said in favor of each of the several types of pumps, viz., piston or plunger pump, gear pump and oil-flooded pump, but we will not attempt to discuss their advantages except very briefly. The plunger pump requires inlet and discharge valves, necessitates some form of packing

and seal and performance is unsatisfactory. The centrifugal pump is the most dependable of the various types, but its capacity and discharge pressure are functions of its speed and the head against which it operates and it does not react smoothly as well as might be expected from a superficial consideration of the problem.

It is thought that a good gear pump driven by the engine offers an satisfactory type for the work on air, and this type is developed by the Army Air Service at McCook Field. This is a type of plunger pump using a system before us as the displacement element and free away with the necessity of packing the driving shaft against leakage. At the present time the cost of this pump is rather high.

Systems Illustrated

A few illustrations showing representative systems will perhaps give a better idea of the accessory pipes, valves and connections than any extended description and may also induce some thought from the reader along the lines of possible and successful improvements. Right now, it may be said that it is believed to be better and safer to induce any system down to the lowest number of essential parts and then make all these parts of proper design and of sufficient strength to

The next obvious step is to embody this return line in the pump and thus do away with the necessary outside piping and fittings. The pump designed for this purpose by G. B. Kirkham is shown in Fig. 3, and the return system using this type of pump is illustrated in Fig. 5.

It will thus be seen from Fig. 3 that such a system is needed in an engine-driven pump embodying the return system, similar to the pump to the stream, and a line from the stream to the pump. In the same way, the return line in the system is used, the main tank is at such an elevation that gas runs from there to the pump by gravity, thus obviating the possibility of air-priming the pump from any slight air leak into the part of the system.

In the system illustrated the only pipe inside the engine compartment of the fuselage is a small length leading from the carburetor bowl directly down to the point where the fuel line enters the fuselage and before this point the line is well outside the fuselage to the pump and streamer mounted below, with the intake line also mounted outside the fuselage. Most of the pipe is therefore removed from any possible source of leakage.

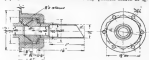


Fig. 3

valves and installed outside where any possible leakage is practically harmless. Furthermore, the chance of the leakage is reduced to a minimum by a reduction of the number of connections necessary, which connections are the potential source of leakage.

Streamers

The matter of pipe connections and fittings which will be the subject of a discussion of its own and will therefore only be dealt with briefly. An essential element of any system is stream of sufficient capacity and of sufficient strength to perform its work without possible trouble. The wire gauze-gasket should not be of less than 30 mesh screen, and an area of 144 to 200 sq. in. per 100 h.p. is not too much. A good mesh laid with a rock at the bottom should be provided for any accumulation of water, and it should be possible to take out the streamer with for cleaning without breaking any of the pipe connections.

Pipe Connections

Annular copper tubing has been almost universally used for connecting engines and the outcome from its long use has been rather poor. This takes care of vibration in the piping, which is a potential source of rupture, but, inasmuch as practice attests rubber, it is always a source of trouble from plastic contraction joints. The common and preferred method is therefore to use a type known as the "Oiler" joint shown in Fig. 5 which consists principally of a metal female inserted between the two ends of the joined pipe, which prevents small pieces of rubber from getting into the system.

Many connections have not heretofore been looked upon with favor, because of their rupture due to vibration. The commercial type of these joints has however not been of proper design to stand the strain. A type of joint, designed by G. B. Kirkham, which has proven satisfactory is shown in Fig. 6. The necessary amount of metal is embedded in this joint and provision is made for proper bracing of the pipe to the fuselage so that the strain of vibration in this direction has been taken care of. It has not been looked, which may mean it to back off, thus opening up the seal and causing a leak.

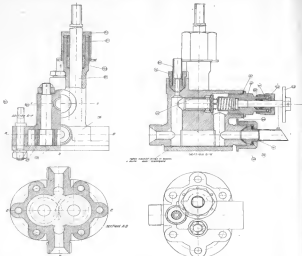


Fig. 4

the engine is idling in a glide, for instance, the pump may be running so fast or faster than when the engine is demanding a maximum quantity of gas, so in level flight or in a climb.

It has happened many times in the Air Mail Service and in other flights that a wind-driven pump has delivered gas to an overboard gravity tank faster than the return line could return it, resulting in much gas being pumped overhead.

Finally, the propeller or compressor is frequently mounted on the driving shaft of the pump and often puts an excessive or other strain on the shaft or other parts of the pump which cause rapid and excessive wear or breakage and also makes

to reduce driving speeds and attracts the one or more plungers at their proper speed and so makes such a pump rather complex.

The gear pump is a satisfactory type when properly designed and constructed of the right materials, considering the small amount of work it is called upon to do in returning gas to fuel tank or several engines. It can be the cause of much trouble when not so designed and built, as many have been in the past. In some pumps the lead carrying the driven gear is carried in the cover plate, making it necessary to accurately draw this plate in place and even then the gears get out of

function correctly, rather than endeavor to provide for all possible misalignments of shaft and so complicate the system.

For instance, instead of providing a complete auxiliary fuel system to be put into operation in case of failure of the main system, it is better to make the main system as it will not fail, and omit the auxiliary system. We do not use two crank shafts or two cam shafts, because one is liable to fail, but we do build the one crank shaft so that its chance of failure is reduced to a minimum.

A single gravity system using a single tank with an arrangement for pumping the engine as starting is shown in Fig. 1. This system is reliable provided sufficient head can be maintained at all times at the carburetor to supply gas under conditions to the engine. A small overflow tank is provided at the upper part of the tank which overflows through small lines into the main tank, and which is used to always supply pressure when maintaining the pump. From this distributor it will be noted that various distributing valves, nozzles and other means provide for about every possible condition of flight. Even this simple system may therefore be somewhat complicated.

Fig. 2 is a diagrammatic illustration of the fuel system used in the original Army Air Service engine. An engine-driven air pump provides sufficient air pressure to drive the motor out of the main tank. An auxiliary tank in the motor wing serves as an emergency tank. A tank pump for pumping air in the motor tank in case of failure of the engine-driven pump, adds another point of apparatus with the necessary piping. A pressure system also adds to the amount of pipe and fittings.

It is to illustrate how a complex system may be simplified in accordance with the thought that reliable simplicity is preferable, reference may be made to Fig. 3. Consider a system in which a pump is used to deliver gas from either side into the main tank through a duplex three-way valve to the carburetor, with a return line through a spring-loaded check valve to either of the two main tanks. The main pump is driven from the engine. A hand pump is attached to the system to furnish gas to the main tank before starting. The obvious first step in simplification is to return the gas from the discharge to the inlet side of the pump through the spring-loaded check valve which will in operation drive gas into the line running to the carburetor. Thus the excess gas pumped, over that required by the engine, is returned to the inlet side of the pump instead of to the main tank.

While most all parts of a plane are fastened by bolts and safety wires, ballast must be bolted onto the proper holding of a motor. In the main, ballast, a sponge rubber is inserted in where. This not only tends to hold the two parts of the motor to a unit, but prevents heating off.

While it is possible to make this motor only in a more or less perfect manner, it is hoped that what has been here presented may lead to a serious consideration of the subject and perhaps to manufacturing all systems and parts that may be found by a thorough trial to be safe and satisfactory for this important part of airplane performance.

Sperry Says Installed Material

The Lawrence Sperry Aircraft Co., Inc., whose experimental material and various models are well known the world over, has recently purchased the entire stock of Aero planes and spare parts belonging to the International Aircraft Corp.

The Sperry Company is marketing these material at prices ranging from \$250 to \$1000 depending on the needs of the customer.

It is understood that Mr. Sperry is bringing out a small airplane which is a modification of the Messinger he built for the Army, which was described in the Nov. 18 issue of this publication and that he will be ready to demonstrate this plane to the public within a month.

J. S. Macdonald, formerly secretary of the International Co., is now with the Sperry Co., in charge of sales.

Club Plans for Royal Aero Club Members

It is now possible for members of the British Royal Aero Club to take up their flights for a solo over the country about Crayke, London, and the other airports or flying clubs. The club is paid for the operation of the craft if the member is not, himself, a pilot. At the standard rate of exchange the cost per hour for an Armstrong 160-horsepower or R.E.C. biplane would be about 65s. A specially "big" single engine runs 10 shillings higher. The rate in British money are 65 and 75 shillings respectively. These charges include engine and oil and insurance of machine and third party in excess of £250.

The conditions for hire are as follows:

1. Flights may be confined to England and within a radius of 150 miles of London, unless special arrangements are made beforehand, and the Member taking the machine are responsible for its return to headquarters.

2. If the machine is away from headquarters for more than two hours the flying time (i.e. time in the air), an additional charge of £15.00 for every 24 hours or part thereof will be made.

3. A machine may not be used for less than half an hour.
4. If a Member uses up his supply of petrol or oil before returning to headquarters, an allowance will be made to him for the extra petrol and oil purchased, on production of receipt for same.

5. Members may carry passengers, but not for hire or reward.

6. A regular pilot who is willing to take up Member's flying machines is kept at the Club.
7. The pilot of the machine must hold an Air Ministry License.

The Club cannot accept responsibility for injury to the Member, or his friends, or property, or for damage to any machine owned by any Member, or injuries to third parties or their property, damage, or loss occasioned by any flight.

All communications as to hiring of machines must be made direct to the Secretary, Royal Aero Club, 2 Clifford Street, London, W.1.

H. A. Brown with Aeromarine

Henry A. Brown, who for the past year and a half was employed by the Manufacturers Aircraft Association in their publicity work, since January 11 has been in charge of advertising and sales for the American Engineering and Sales Co., Inc., Tower Building, New York.

Third Annual Aviators Ball

The Third Annual Aviators Ball will be held this year at the Waldorf Astoria on Thursday, April 7, 1931. Owing to the unprecedented success of the Ball, held in 1929 and 1930, when the capacity of the Ritz-Carlton was greatly exceeded, the Ball Committee has engaged the entire second floor of the Waldorf, which allows half as much more dancing space for the coming Ball. The tickets will cost the same as in past years—five dollars plus fifty cents war tax per person.

The four Committees of Patronage, Patron, Patroness and Aviator will have chosen the names of the members of their committee will be just as representative as on the previous memorable occasions. All communications should be addressed to the Ball Committee, Headquarters, Aero Club of America, 11 East 55th St., New York.

Long Distance Flying

The following article, setting forth the importance of long distance flying, is of special significance as it comes from Captain Howard Douglas, A. S., the Publisher of the Atlantic Expedition—K.

Since the close of the war, the Air Service of every important country in the world has been conducting long distance flights. One of the first and one of the most spectacular was the crossing of the Atlantic Ocean in a non-stop flight, and also of supreme value was the splendid performance of the U. S. Navy in crossing the Atlantic in a seaplane. The British air force has crossed England and Atlantic and across to Cape Town. The Alps have been explored by airplanes and flights from France to Northern Africa have been carried out. In the United States, the Army Air Service has demonstrated a military force, transportation of troops, a flight from San Diego, California to Jacksonville, Florida, a flight across the rim of the United States of almost 30,000 miles, and a flight from New York City to Nome on the Bering Sea and return, a distance of approximately 9,000 miles.

There are still individuals, presumably of average intelligence, who will ask the question, "Why are these dangerous flights attempted?" It is a matter of courtesy to explain the reasons, back to the very beginning, why we have paid so much attention to aviation and its development, the reason is very obvious.

It is only the tendency to accomplish difficult things that will allow aviation, long distance flying, to be so successful, and with such unequalled success during the past year that it is possible to foresee the time when no part of the world will be inaccessible to airplanes. It has not been the policy of the Air Service of any country to fly for the sake of flying, when airplanes, specially designed for long distance flying, were built. Using the equipment we have had at hand for war purposes, we have successfully accomplished flights that a few years ago were considered impossible. We have carried flying operations into parts of the world where airplanes have never been before. On the Atlantic trip, there was money transferred which no man had ever seen before. The extraordinary value of such flights is thus established. It demands a daring imagination to even estimate the extent and future possibilities of aerial exploration.

Aside from the exploratory value of long distance reconnaissance flights, there is every national and military value. Every eye and tongue, regardless of attainment by any other means, is to be had. Pilots and mechanics, who participate in such flights, carry on operations thousands of miles from their headquarters. Every long distance flight ever attempted has given the flying personnel a wonderful opportunity to study the art of flying. It affords the best opportunity to check the value of navigating instruments. Every flight has been a school of practical instruction in navigation, in the supply and navigation. The mechanical solution of all these problems is the best possible training for Air Service personnel. Practice flying around an airplane is essential, but however important such flying may be, it cannot be carried on in any way with long distance cross country flying. These flights already successfully carried out have done more than any one thing to foster and develop commercial aviation, and from a military and naval point of view, the experience and training gained have been invaluable.

U. S. Commercial Aircraft Companies

Prepared by the Manufacturers Aircraft Association

After careful compilation of Air Service manufacturers and operators reports, the Manufacturers Aircraft Association estimated that there were on Jan. 1, 1931, one thousand aircraft in commercial use in the United States and Canada.

Due to the unusual nature of much of the flying, and the lack of any federal system of registration it was impossible to trace the activities of all these machines. But numerous reports were received from eighty-eight companies, con-

taining nearly five hundred machines. On this basis it is estimated that the one thousand aircraft figure is within miles and carried approximately 250,000 passengers. The Manufacturers Aircraft Association reports that during 1930 the Army Air Service flew 6,250,000 miles, Naval Aviation 1,000,000 miles, and the Air Mail 1,000,000 miles. The total mileage in the air is estimated at 85,250,000.

COMMERCIAL AIRCRAFT OPERATING COMPANIES IN THE UNITED STATES AND CANADA

Name of Company	Address	Address	Year of Operation	Operating Aircraft	Passenger Capacity	Freight Capacity	Other Notes
1. Aerial Trans Co.	Seattle, Wash.	1 R. E. L. F.	1928	100	1,000	0	2,500
2. Aeromarine Engineering & Service Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
3. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
4. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
5. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
6. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
7. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
8. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
9. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
10. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
11. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
12. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
13. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
14. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
15. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
16. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
17. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
18. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
19. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
20. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
21. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
22. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
23. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
24. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
25. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
26. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
27. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
28. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
29. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
30. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
31. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
32. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
33. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
34. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
35. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
36. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
37. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
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41. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
42. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
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47. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
48. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
49. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
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51. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
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57. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
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98. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
99. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500
100. Aero-Naval Co.	San Francisco, Cal.	1 R. E. L. F.	1928	100	1,000	0	2,500

United States Civil-Service Examination

Aeronautical Draftsman, National Advisory Committee for Aeronautics

The United States Civil Service Commission announces an open competitive examination (No. 127) for aeronautical draftsmen. Testing is at the Langley Research Laboratory, Langley Field, Hampton, Va., at the salaries indicated below, and in positions requiring similar qualifications, which will be filled from this examination, unless it is found to the interest of the service to fill any vacancy by promotion, transfer, or reversion.

Citizenship and sex.—All citizens of the United States who meet the requirements, both educational and technical, for the examination, appearing there, however, have the legal right to specify the sex desired in requesting consideration of candidates.

Form of entrance.—Candidates will not be required to report for examination at any place, but will be rated upon their education and experience as shown by the seven state grades in their applications and corroborative evidence.

Grades and experience.—The register of grades will be divided into four grades, the requirements, duties, and maximum requirements for eligibility for each grade being as stated below. The compensation, with the range thereof for each grade will depend upon the qualifications of the applicant as shown in the examination.

Grade 1, \$1,500 to \$1,600 a Year

Duties.—The duties of this grade require the services of draftsmen for tracing and for some detailing from aircraft or engine layouts.

Requirements.—Applicants must show:

- That they have completed a common-school education and have had four years' experience as copyist or apprentice draftsman, or as one of such as results of such experience, applicants may substitute such successfully completed year at high school or as a mechanical or electrical engineering course in a college or university of recognized standing; and
- Except in the case of graduates in mechanical or electrical engineering, at least six months of the mechanical or electrical experience must have been spent in copying and detailing from aircraft or high-speed internal-combustion engine drawings, or mechanical assembly drawings.

Grade 2, \$1,600 to \$1,800 a Year

Duties.—The duties of this grade require the services of draftsmen for detailing from aircraft or engine layouts.

Education and experience.—Applicants must show that they possess the following qualifications:

- A good common-school education and four years' experience as mechanical draftsman; or in lieu of such six months of such experience applicants may substitute such successfully completed year at high school or as a mechanical or electrical engineering course in a college or university of recognized standing; and

(b) One year's experience as mechanical draftsman, performing detailing from aircraft or high-speed internal-combustion engine drawings or mechanical assembly drawings, or as one-half year's experience in general layout and detailing of special machinery or machine tools.

Grade 3, \$1,800 to \$1,900 a Year

Duties.—The duties of this grade require the services of draftsmen for detailing and for some work in assisting on the preparation of notes, checking plans and specifications, making sketches and drafting engine layouts.

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Age.—Applicants must have reached their eighteenth birthday before the retirement act, the date of the examination. In view of the retirement act, should the appointing officer or regular certificate will not be made of applicants who have reached their fifty-fifth birthday.

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Applications.—Applicants should at once apply for Form 1232, stating the title of the examination desired, to the Civil Service Commission, Washington, D. C.; the Secretary of the United States Civil Service Board, Cantonhouse, Boston, Mass.; New York, N. Y.; New Orleans, La.; Honolulu, Hawaii; Port Office, Philadelphia, Pa.; Atlanta, Ga.; Cincinnati, Ohio; Chicago, Ill.; St. Paul, Minn.; Seattle, Wash.; San Francisco, Cal.; Old Cantonhouse, St. Louis, Mo.; Administration Building, Rabon Heights, Canal Zone; or to the Chairman of the Porto Rico Civil Service Commission, San Juan, P. R.

Applicants should be properly educated, standing, the medical and county officer's certificate, and must be filed with the Civil Service Commission, Washington, D. C., prior to the last of June business on March 22, 1921.

The last title of the examination, as given at the head of the announcement, should be stated in the application form. **Preference.**—Applicants entitled to preference should at least show experience in the mechanical engine, or as one of such as results of such experience in detailing of aircraft or high-speed internal-combustion engine drawings, or as one-half year's experience in general layout and detailing of special machinery or machine tools.

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The following have courses in preparation: Georgia School of Technology, Atlanta, Ga.; Iowa State College, Ames, Ia.; Ohio State University, Columbus, Ohio.; State University of Oklahoma, Norman, Okla.; University of California, Berkeley, Cal.; Stanford University, Stanford, Cal.; University of Wisconsin, Madison, Wis.; University of Illinois, Urbana, Ill.; Worcester Polytechnic Institute, Worcester, Mass.



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